

In the Abstract

Please replace the Abstract as presented in the underlying International Patent Application No. PCT/DE2005/000329 with the following amended Abstract:

ABSTRACT

~~The invention relates to a~~ A gas turbine vane, especially a vane pertaining to an aircraft engine, comprising a blade-(11) and a vane footing-(12). ~~Said~~ The blade-(11) is defined by a flow inlet edge or a front edge-(13), a flow outlet edge or a rear edge-(15), and a blade surface-(15) extending between the front edge-(13) and the rear edge-(14) and forming a suction side-(16) and a pressure side-(17). ~~According to the invention, t~~The suction side-(17) of the blade-(11) comprises includes at least one microprofiled or microstructured region-(18; 20, 21, 22) for optimizing the flow around the blade-(11).